



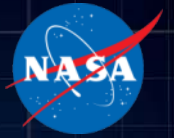
Exploration Operational Concepts

Microbiology Water Forum

Craig E. Kundrot, PhD
July 27, 2011

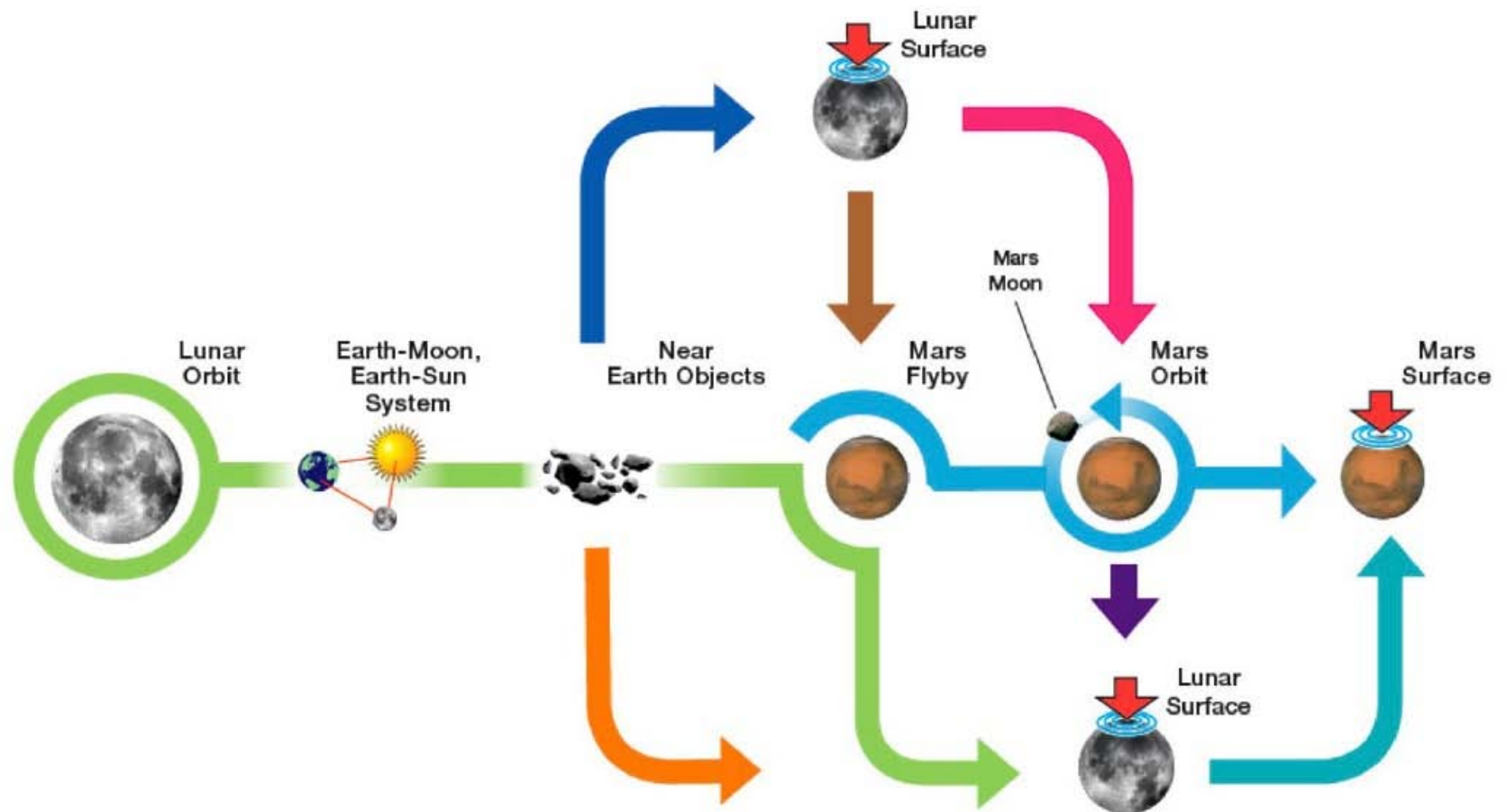
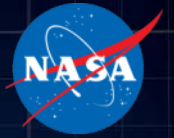


Where to Next?

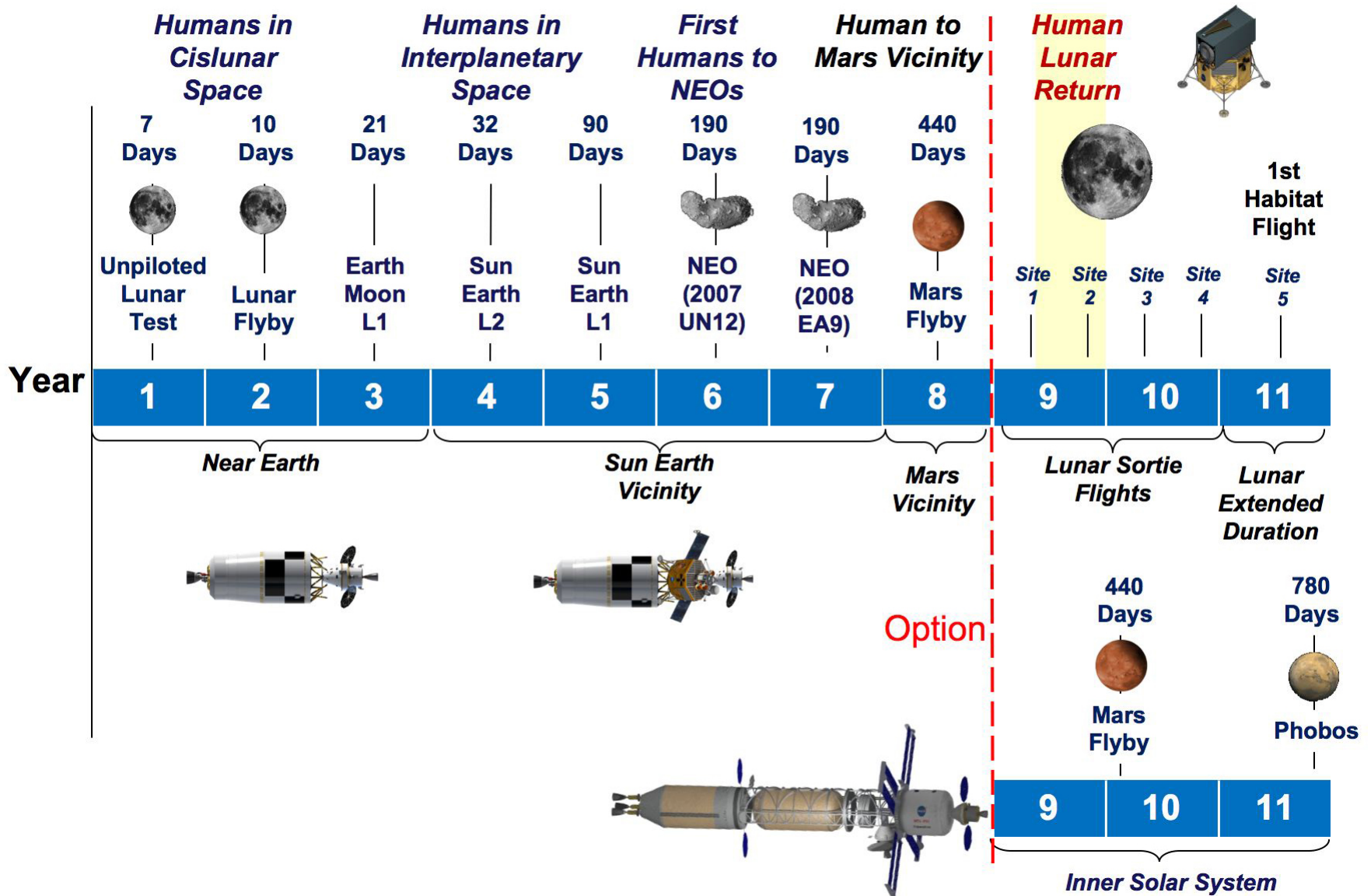
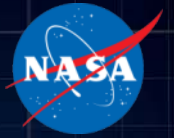


- October 2009
- Delivered to NASA Administrator and Office of Science and Technology Policy
- Provides the context for discussing future destinations

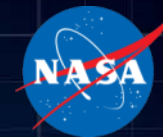
The Flexible Path



One Path with a Fork



Notional Architecture Elements



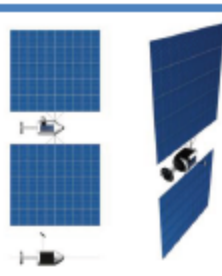
Space Launch
System (SLS)-HLLV



Multi-purpose
Crew Vehicle
(MPCV)



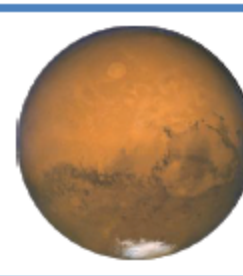
Cryogenic
Propulsion Stage
(CPS)



Solar Electric
Propulsion (SEP)



Lander

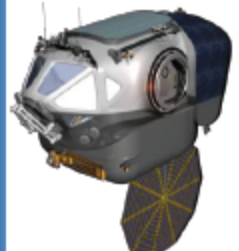


Mars Elements

Graphics are Notional Only – Design and Analysis On-going



EVA Suit



Multi-Mission Space
Exploration Vehicle
(MMSEV)



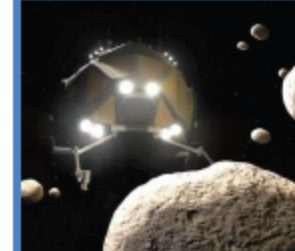
Deep Space Habitat
(DSH)



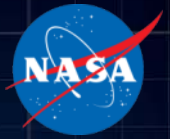
Robotics & EVA
Module (REM)



Kick Stage



NEA Science Package



Key Technical Architecture Observations To Date



- ◆ Advanced in-space propulsion (e.g., solar electric propulsion {SEP}) is a big enabler: Reduces launch mass by 50% (factor of 2) and mass growth sensitivity by 60%
- ◆ A balance of ELVs and HLLVs is optimal for varying mission needs
- ◆ Shuttle-derived HLLV option (100t-class evolvable to ~130t for deep space, full capability missions) meets more current FOMS than other options, although out-year affordability is still a fundamental challenge for long term exploration. Alternative design analysis continues to be part of NASA's strategy, coupled with an assessment of possible affordability initiatives.
- ◆ HLLV and crew vehicle should be a human-rated system
- ◆ ELV-only solution not optimal given all factors
- ◆ Staging at HEO or Earth-Moon L1 for deep space missions better than LEO
- ◆ Crew Transportation Vehicle (CTV) full ascent and entry capability is needed
- ◆ Additional capability, such as the MMSEV needed for EVA and robotics capability
- ◆ High reliability ECLSS is desired over fully closed loop ECLSS except for Mars missions
- ◆ In-Situ Resource Utilization (ISRU) is an enabler, particularly for surface missions
- ◆ Modularity and commonality aid key affordability FOM

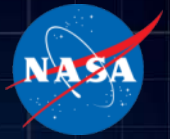
• HLLV=Heavy Lift Launch Vehicle
• CTV=Crew Transportation Vehicle
• MMSEV=Multi-mission Space Exploration Vehicle

• EVA=Extravehicular Activity
• SEP=Solar Electric Propulsion
• ECLSS=Environmental Control and Life Support Systems

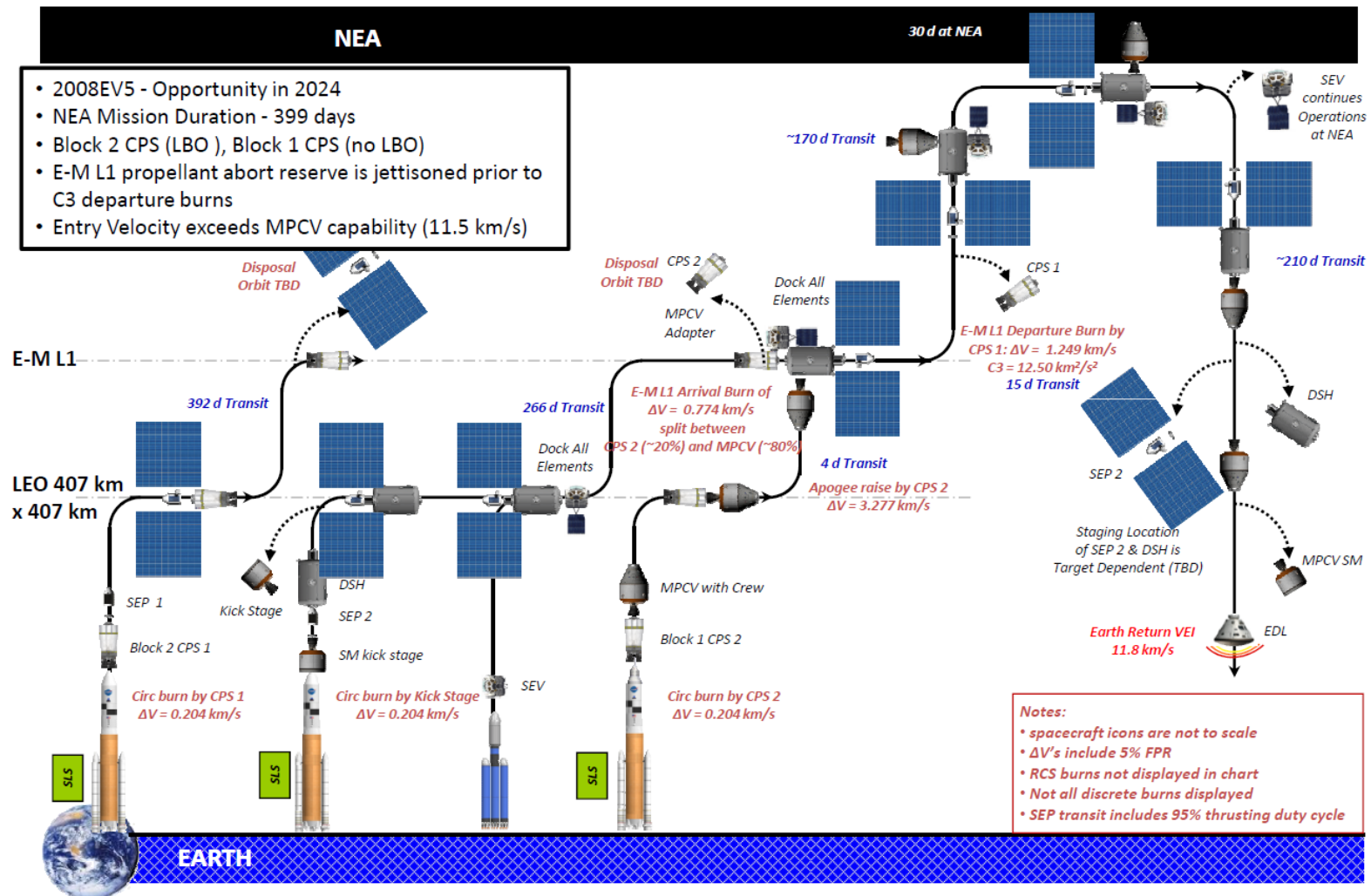
For Public Release

14

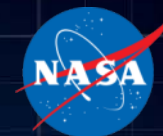
A Sequence for an Asteroid



Reference NEA Mission: DRM 34B (NEA 2008EV5 with SEP)



Technology Applicability to Destination (1)



	LEO (31A)	Adv. LEO (31B)	Cis-Lunar (32A,B & 33A,B)	Lunar Surface - Sortie (33C)	Lunar Surface - GPOD (33X)	Min NEA (34A)	Full NEA (34B)	Mars Orbit	Mars Moons (35A)	Mars Surface (35B)
LO2/LH2 reduced boiloff flight demo										
LO2/LH2 reduced boiloff & other CPS tech development										
LO2/LH2 Zero boiloff tech development										
In-Space Cryo Prop Transfer										
Energy Storage										
Electrolysis for Life Support (part of Energy Storage)										
Fire Prevention, Detection & Suppression (for 8 psi)										
Environmental Monitoring and Control										
High Reliability Life Support Systems										
Closed-Loop, High Reliability, Life Support Systems										
Proximity Communications										
In-Space Timing and Navigation for Autonomy										
High Data Rate Forward Link (Ground & Flight)										
Hybrid RF/Optical Terminal (Communications)										
Behavioral Health										
Optimized Exercise Countermeasures Hardware										
Human Factors and Habitability										
Long Duration Medical										
Biomedical countermeasures										
Space Radiation Protection – Galactic Cosmic Rays (GCR)										
Space Radiation Protection – Solar Proton Events (SPE)										
Space Radiation Shielding – GCR & SPE										
Vehicle Systems Mgmt										
Crew Autonomy										
Mission Control Autonomy										
Common Avionics										
Advanced Software Development/Tools										
Thermal Management (e.g., Fusible Heat Sinks)										
Mechanisms for Long Duration, Deep Space Missions										
Lightweight Structures and Materials (HLLV)										
Lightweight Structures and Materials (In-Space Elements)										

A Sequence for Mars

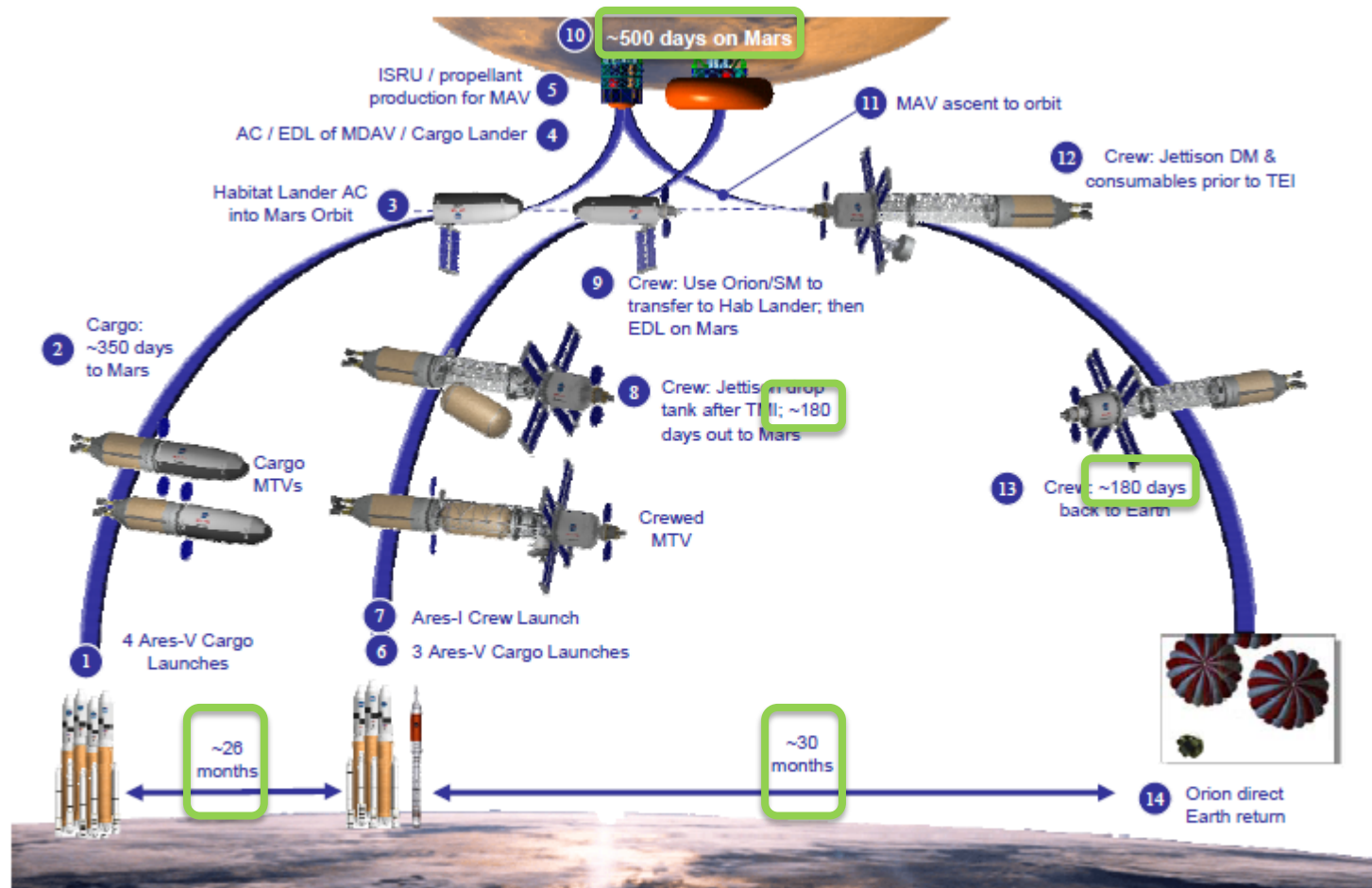
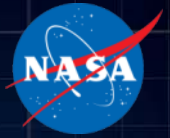
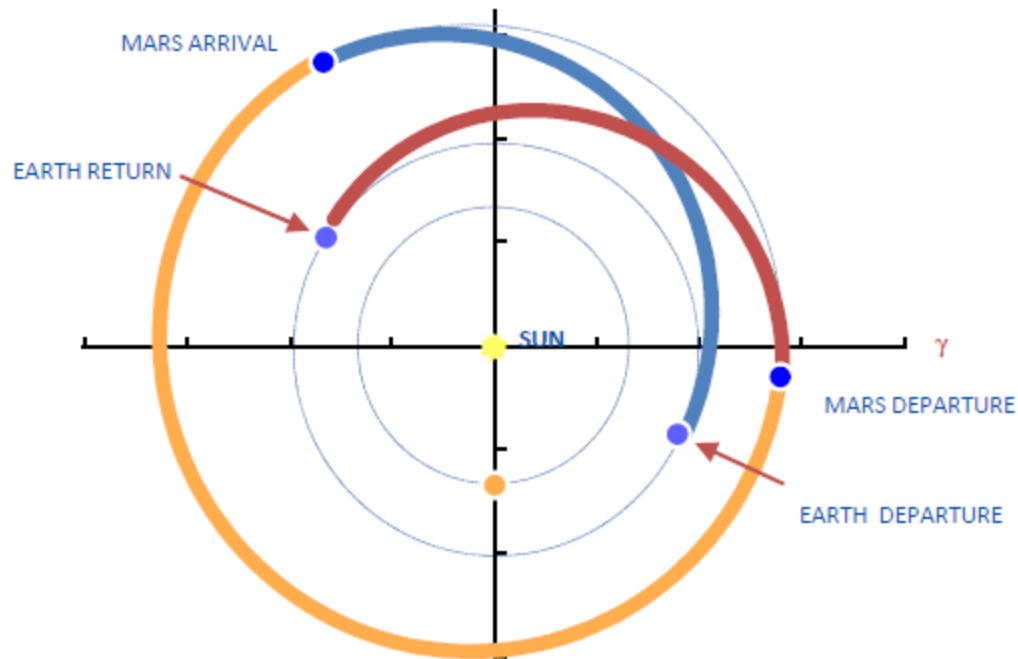
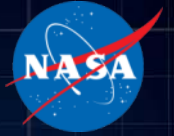


Figure 2-2. Mars Design Reference Architecture 5.0 mission sequence summary (NTR reference).

Mars Mission Length



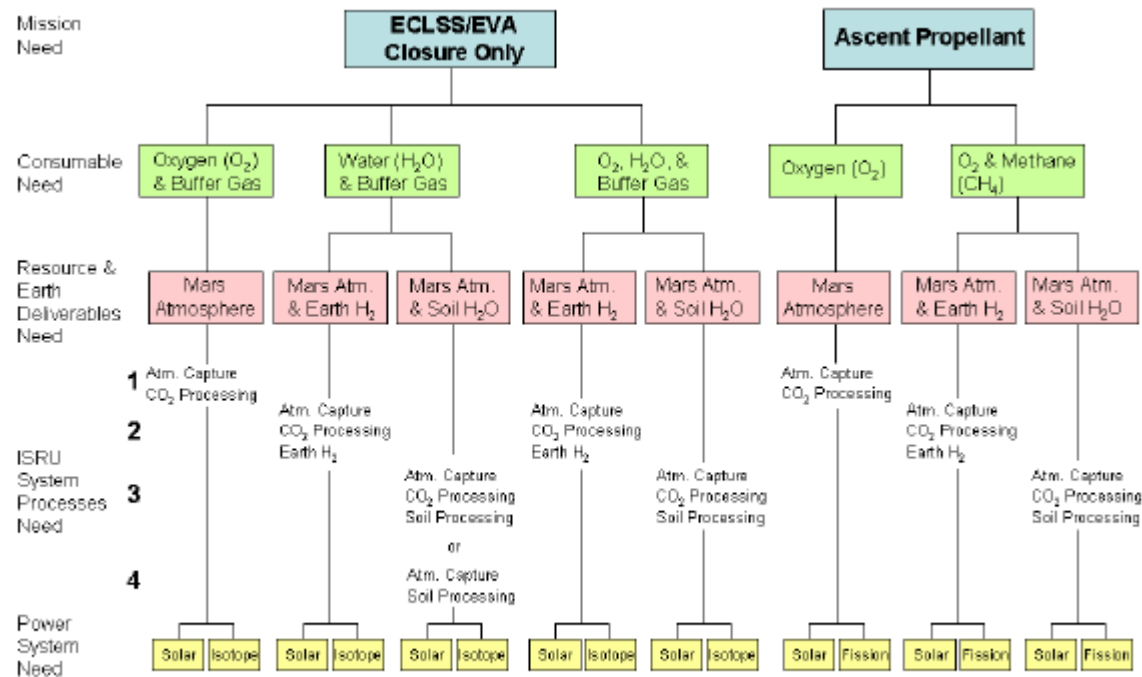
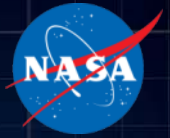
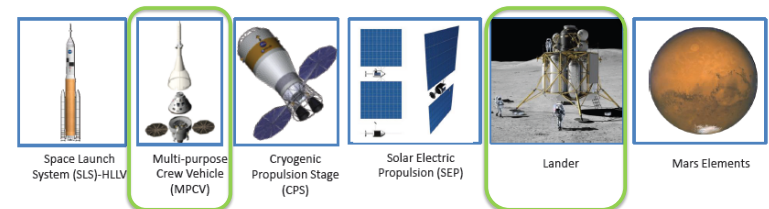


Figure 6-5. ISRU trade tree.

Summary



- Systems
 - Capsule
 - (Lander)
 - EVA Suit
 - Exploration Vehicle
 - Deep Space Habitat
- Mission Duration
- ISRU on Mars
- Fully closed loop ECLSS for Mars



Graphics are Notional Only – Design and Analysis On-going

